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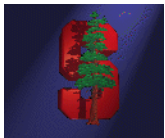
**AA284a**  
**Advanced Rocket Propulsion**

**Lecture 3**  
**Review of Thermodynamics and Chemistry**

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**Fall 2019**



**Stanford University**



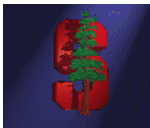
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# Advanced Rocket Propulsion

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## Review of Thermodynamics-Definitions

- System: A group of entities distinguished from its surroundings
  - Mass and energy transfer is allowed
  - Change of volume is allowed
  - Example: Human body
- Transfer quantities
  - Heat (transfer to the system):  $\delta Q$
  - Work (done by the system):  $\delta W$
  - Mass transfer:  $\delta M$
- Definitions:
  - Open System: Mass transfer allowed
  - Closed System: Mass transfer **not** allowed
  - Adiabatic System: Heat transfer is **not** allowed



Reference on Thermodynamics: I. Klotz and R. Rosenberg, "Chemical Thermodynamics"  
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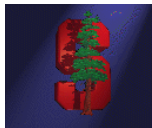
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## Review of Thermodynamics-Definitions

- State of the system: Quantify the status of the system
  - Extrinsic properties: E, H, G, M, N
  - Intrinsic properties: T, P, z, K
  - Extrinsic properties can be converted to intrinsic properties
    - Example:  $e=E/M$
- Change of State:
  - Potential drives the system to change
  - When the potential diminishes system reaches an equilibrium.
- Mission of Thermodynamics
  - Governs the rules of change of state
  - Move from state A to state B is feasible or not for specified heat and mass transfer and work done
  - Thermodynamics does not answer the questions how fast or what is the exact form of the process.



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## Review of Thermodynamics-Laws

- **Zeroth Law:**

- If A is in equilibrium with C and B is in equilibrium with C, than A and B must be in equilibrium.

- **First Law:**

- Conservation of energy
- Change in the internal energy must be equal to the heat added to the system minus the work done by the system

$$d e = \delta q - \delta w$$

- For simple materials only work is the pressure work.

$$\delta w = P dv$$

- In terms of enthalpy

$$d h = \delta q + v dP$$

*e* : Internal Energy

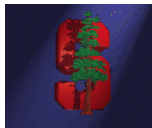
*h* : Enthalpy

*q* : Heat Transfer

*w* : Work Done

*P* : Pressure

*v* : Specific Volume



# Advanced Rocket Propulsion

## Review of Thermodynamics-Laws

- **Second Law:**

- Reversible/Irreversible processes
  - Imagine a time dependent physical process governed by a set of equations
  - If these equations are invariant with regard to the sign of the time variable the process is reversible, else irreversible
- There exists a system variable, entropy, with the following definition for a single component system

$$d s = \frac{\delta q}{T} \quad d S = \frac{\delta Q}{T}$$

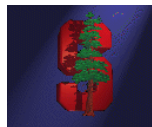
- Explicit property
- Second law of thermodynamics

$$d S_{universe} \geq 0$$

*S : Entropy*

*s : Specific Entropy*

*T : Temperature*



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## Review of Thermodynamics-Laws

- **Second Law:**

- Or

$$d S_{system} + d S_{surroundings} \geq 0$$

- For an isolated system

$$d S_{system} \geq 0$$

- Entropy is a measure of “disorder” or lack of “information” on the possible microstates

- Boltzmann’s Equation:  $S = k \ln(N_{micro})$

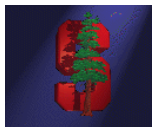
- All real processes are irreversible

*k : Boltzmann's Cons.*

*N<sub>mic</sub> : Number of Micro States*

- **Third Law:**

- Planck’s Formulation: Value of entropy of a pure liquid or solid approaches zero at 0 K

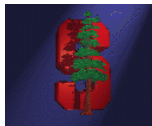


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## Review of Thermodynamics-Cycles

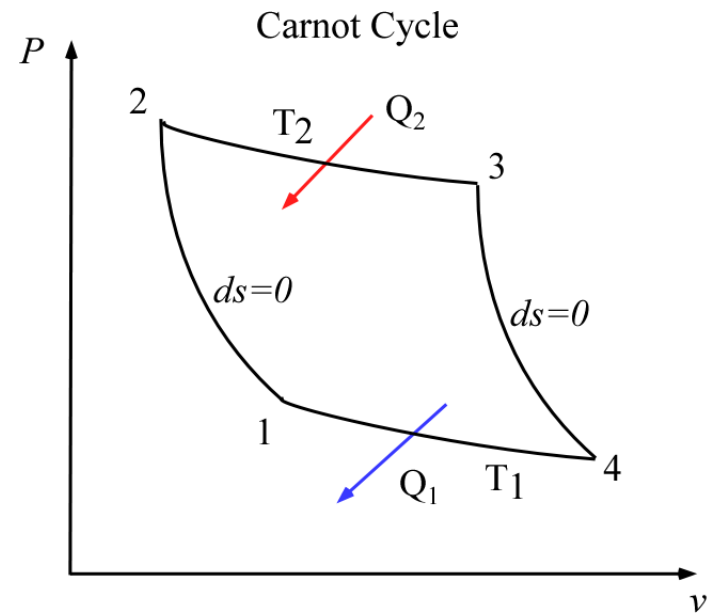
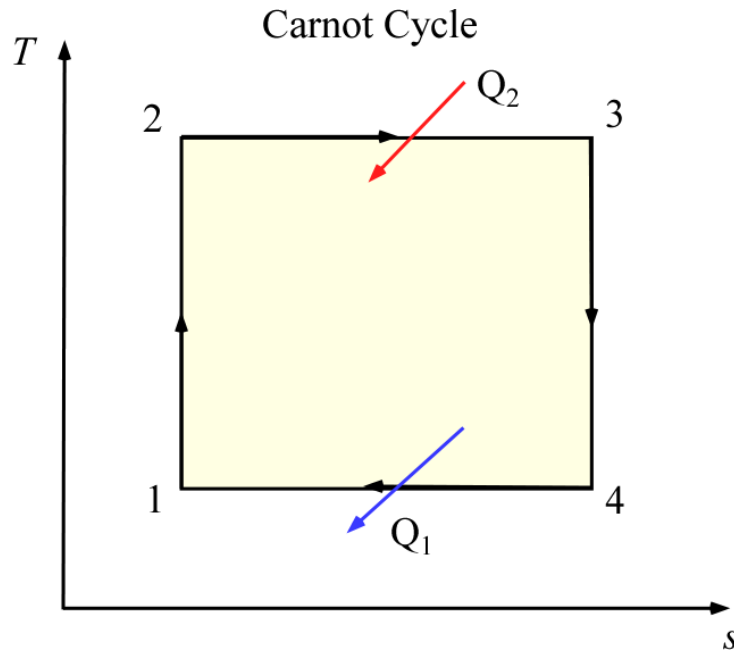
- Heat machines convert thermal energy into mechanical energy according to the laws of thermodynamics.
- Many machines work in cycles. Working fluid returns to the original state that it started.
- Carnot Cycle:
  - Two constant temperature heat transfer processes and two isentropic compression expansion processes
  - Carnot cycle efficiency: 
$$\eta_C = 1 - \frac{Q_1}{Q_2} = 1 - \frac{T_1}{T_3}$$
- Bryton Cycle:
  - Two constant pressure heat transfer processes and two isentropic compression expansion processes
  - Bryton cycle efficiency: 
$$\eta_B = 1 - \frac{\Delta h_{cooling}}{\Delta h_{heating}} = 1 - \frac{T_1}{T_2} < 1 - \frac{T_1}{T_3}$$
- Cycle efficiency increases with increasing temperature ratio
- Carnot cycle is always the best efficiency heat machine operating between two specified temperature extremes ( $T_1$  and  $T_3$ ).

$\eta$  : Efficiency

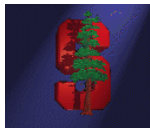


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## Review of Thermodynamics-Carnot Cycle



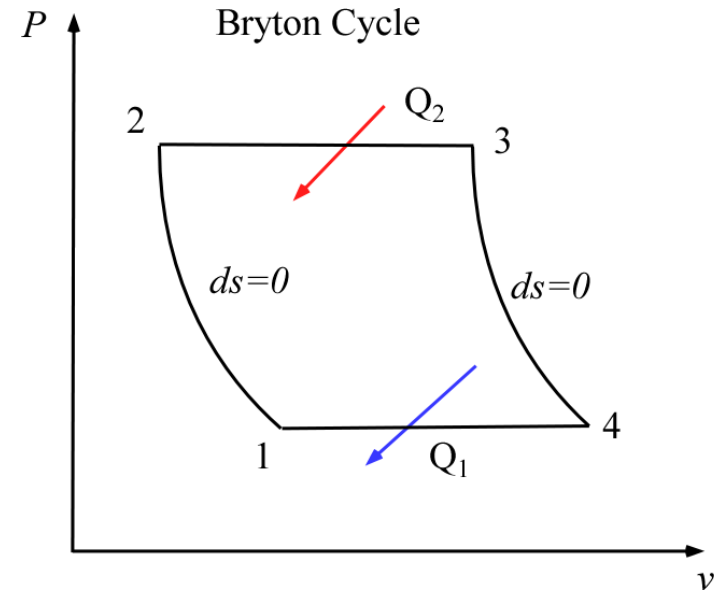
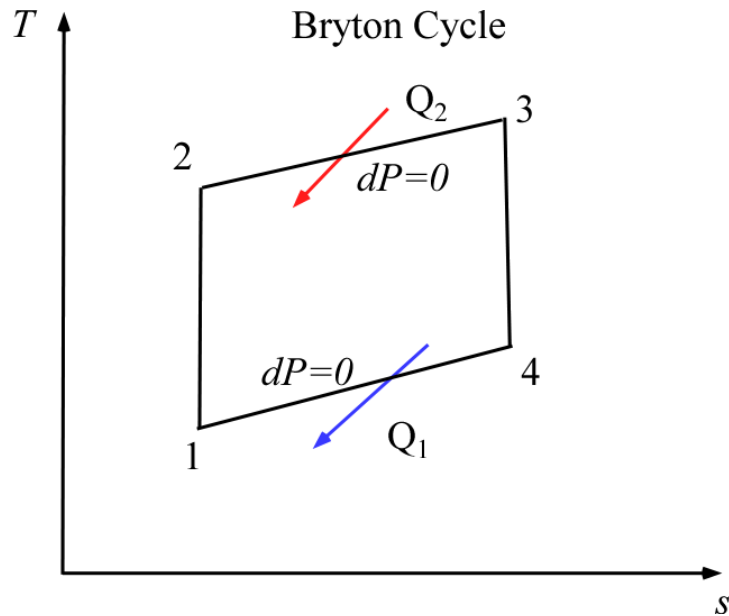
- Carnot Cycle Efficiency  $\eta_C = \frac{W}{Q_2} = \frac{Q_2 - Q_1}{Q_2} = 1 - \frac{Q_1}{Q_2}$
- Isothermal/Isentropic Branches:  $s_1 - s_4 = -\frac{Q_1}{T_1}$   $s_3 - s_2 = \frac{Q_2}{T_2}$   $s_2 = s_1$   $s_3 = s_4$
- Combine:  $\frac{Q_2}{T_2} = \frac{Q_1}{T_1} \implies \eta_C = 1 - \frac{T_1}{T_2} = 1 - \frac{T_1}{T_3} = 1 - \frac{T_{cold}}{T_{hot}}$





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## Review of Thermodynamics - Bryton Cycle

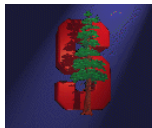


- Bryton Cycle Efficiency  $\eta_B = 1 - \frac{Q_1}{Q_2} = 1 - \frac{h_4 - h_1}{h_3 - h_2}$

- Perfect Gas:  $\eta_B = 1 - \frac{T_4 - T_1}{T_3 - T_2}$

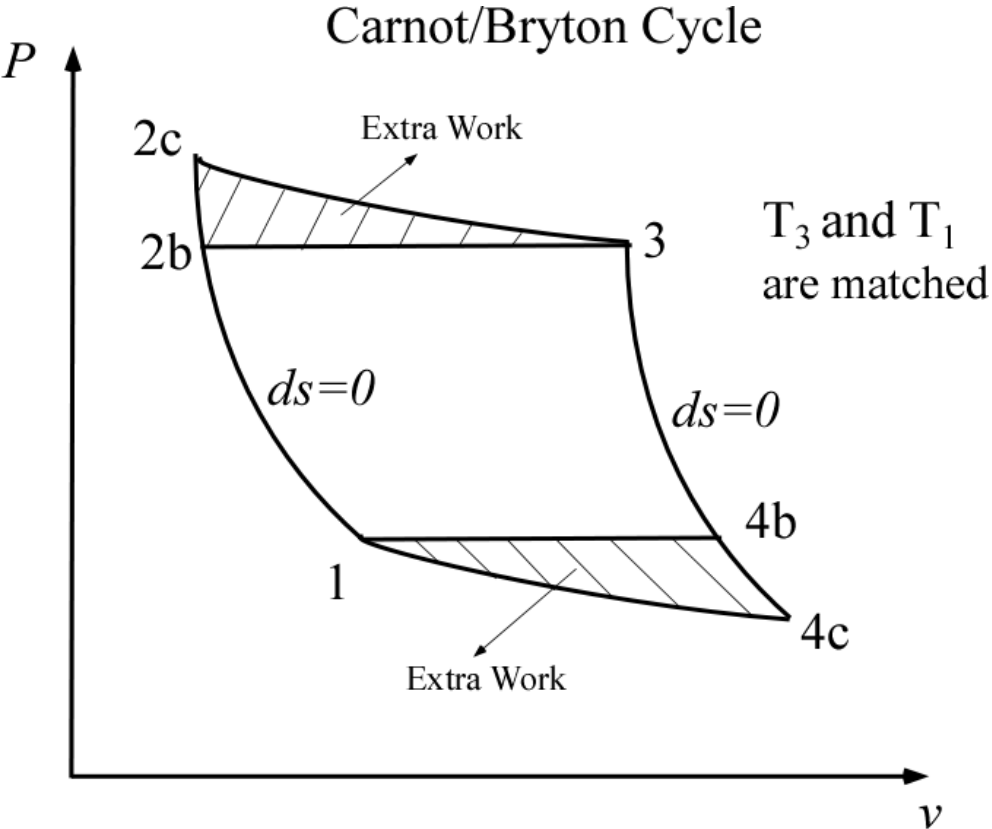
- From Isentropic and Isobaric Branches:  $\frac{T_4}{T_3} = \frac{T_1}{T_2}$

$$\Rightarrow \eta_B = 1 - \frac{T_1}{T_2} \quad \text{but} \quad T_3 = T_{hot} > T_2$$



# Advanced Rocket Propulsion

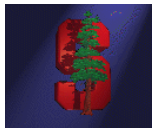
## Review of Thermodynamics – Cycle Comparison



- Work done by the fluid

$$W = \oint P dv$$

- For the same extreme temperatures the Carnot cycle is more efficient than the Bryton cycle
- This conclusion is valid for all other cycles.
- Thus Carnot cycle sets the upper limit for the efficiency of a heat engine operating at two set temperatures
- Nonideal behavior is due to
  - Non-isothermal heat transfer
  - Non-isentropic expansion and compression



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## State Equation

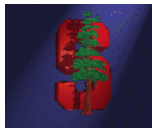
- State equation relates the state variables of a substance to each other
- Gibb's Phase Rule:

$$\text{Number of Phases} + \text{Independent Intensive Properties} = 2 + \text{Number of Components}$$

$$( P + V = C + 2 )$$

### – Examples

- If  $P=1$  and  $C=1$ ,  $V=2$  (One phase one component)
- If  $P=2$  and  $C=1$ ,  $V=1$  (Two phase one component)
- If  $P=1$  and  $C=2$ ,  $V=3$  (One phase two component)



# Advanced Rocket Propulsion

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## State Equation

- State Equation for a one component system:

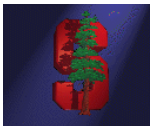
$$f(P, v, T) = 0 \qquad P = P(v, T)$$

- Examples
  - Ideal Gas Equation:

$$P = \rho R T$$

- Ideal Liquid Equation:

$$\rho = \text{cons.}$$



# Advanced Rocket Propulsion

## Real Gases

- Ideal gas approximation fails at high pressures/low temperatures
  - Volume of the gas molecules become non trivial
  - Attraction/repulsion forces between molecules become significant
- Real gas. Compressibility factor (read z from charts)

$$z \equiv \frac{P}{\rho RT} = \frac{PV_m}{R_u T}$$

$V_m$  : Molar Volume

- Equation of State (EOS) for real gases (Cubic equations)
  - Van der Waals

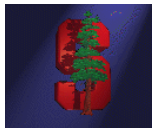
$$\left( P + \frac{a}{V_m^2} \right) (V_m - b) = R_u T$$

- Redlich-Kwong EOS

$$\left( P + \frac{a}{T^{0.5} V_m (V_m - b)} \right) (V_m - b) = R_u T \quad a = \frac{0.42748 R^2 T_c^{2.5}}{P_c} \quad b = \frac{0.0866 R T_c}{P_c} \quad z_c = \frac{P_c V_m}{R_u T_c} = \frac{1}{3}$$

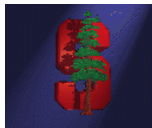
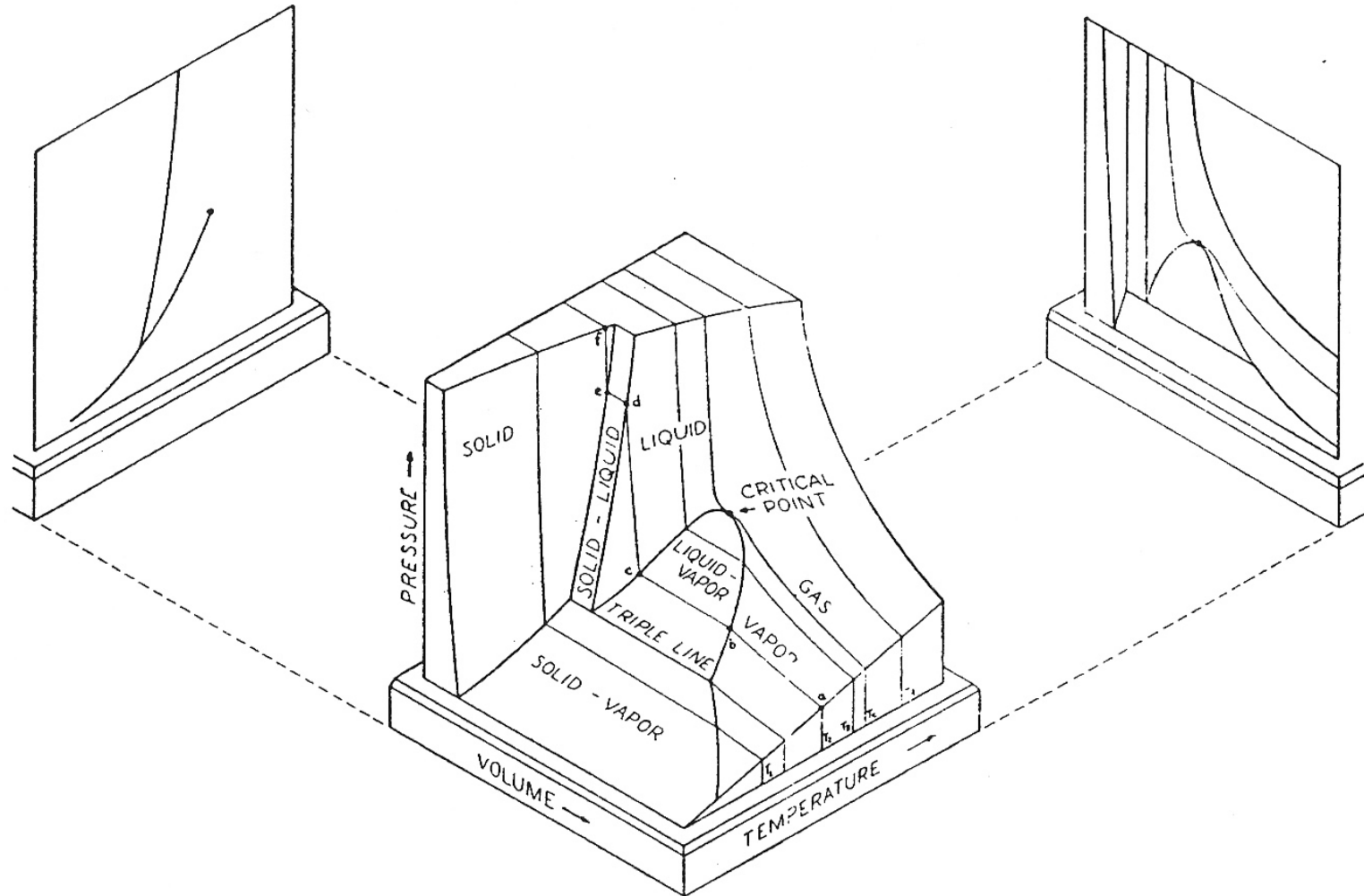
- Peng-Robinson EOS

$$P = \frac{R_u T}{V_m - b} - \frac{a(T)}{V_m (V_m + b) + b (V_m - b)}$$



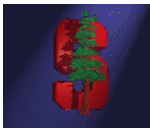
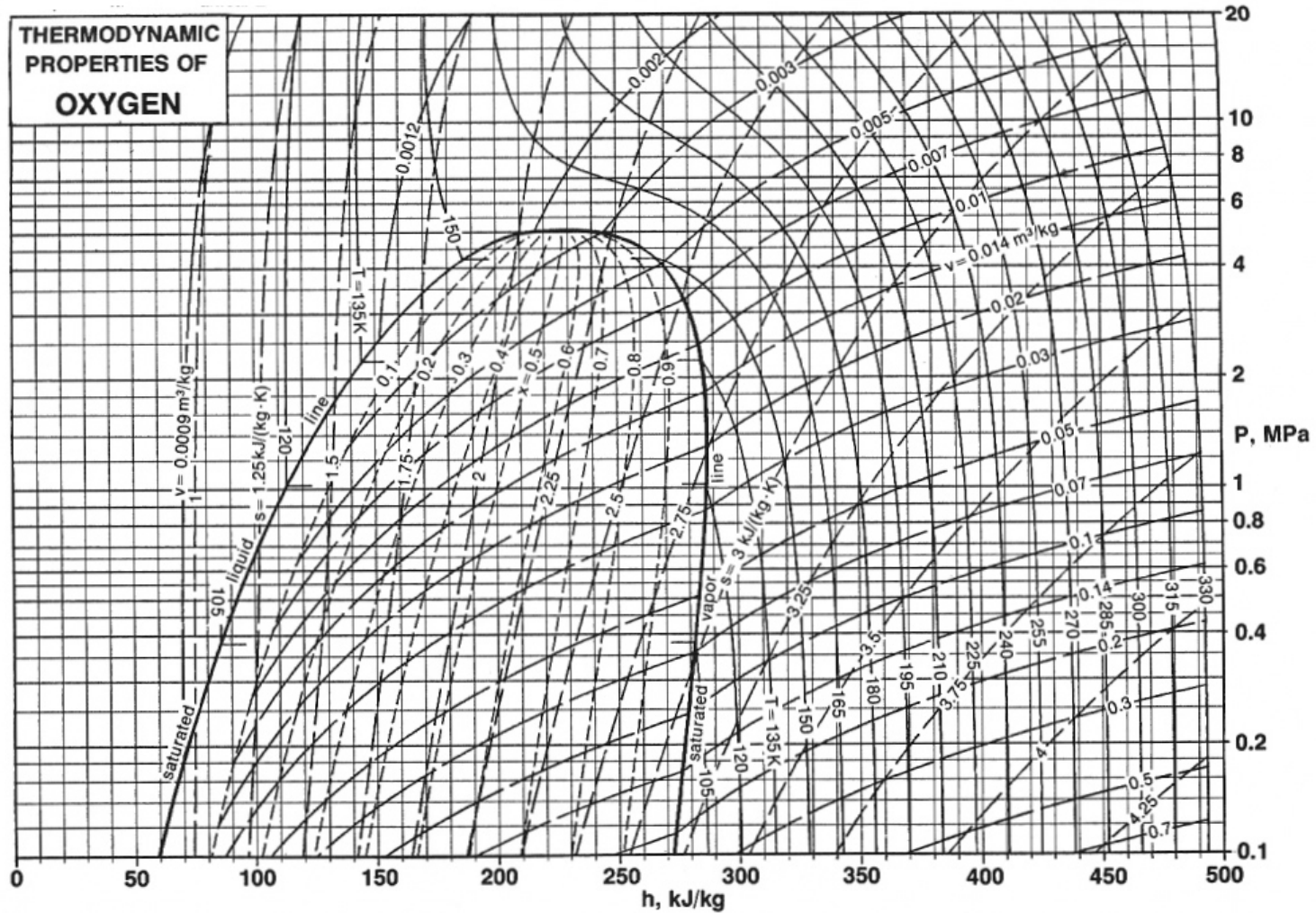
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## P-V-T Diagram of a Single Component Substance



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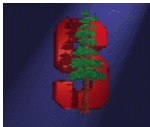
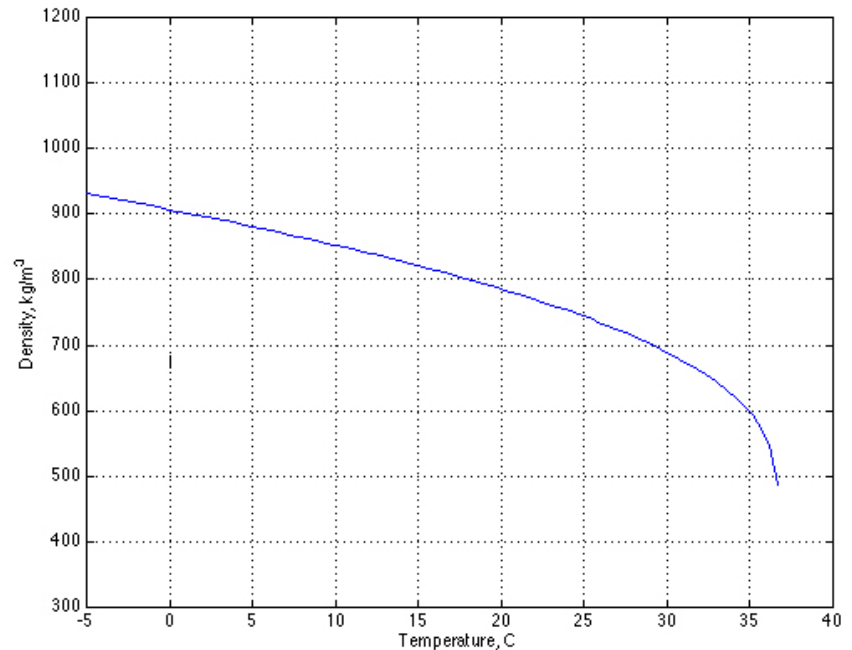
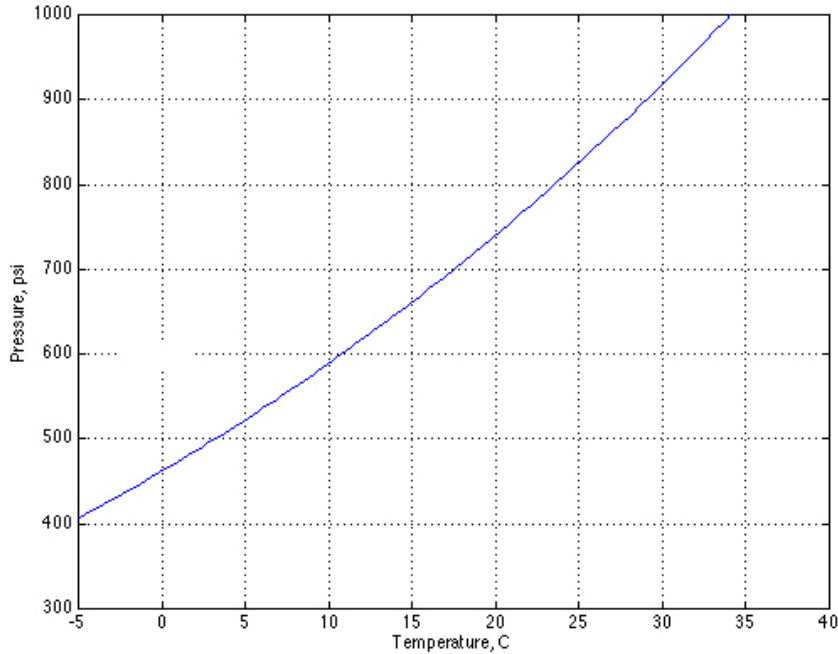
## P-h Diagram Molecular Oxygen





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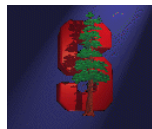
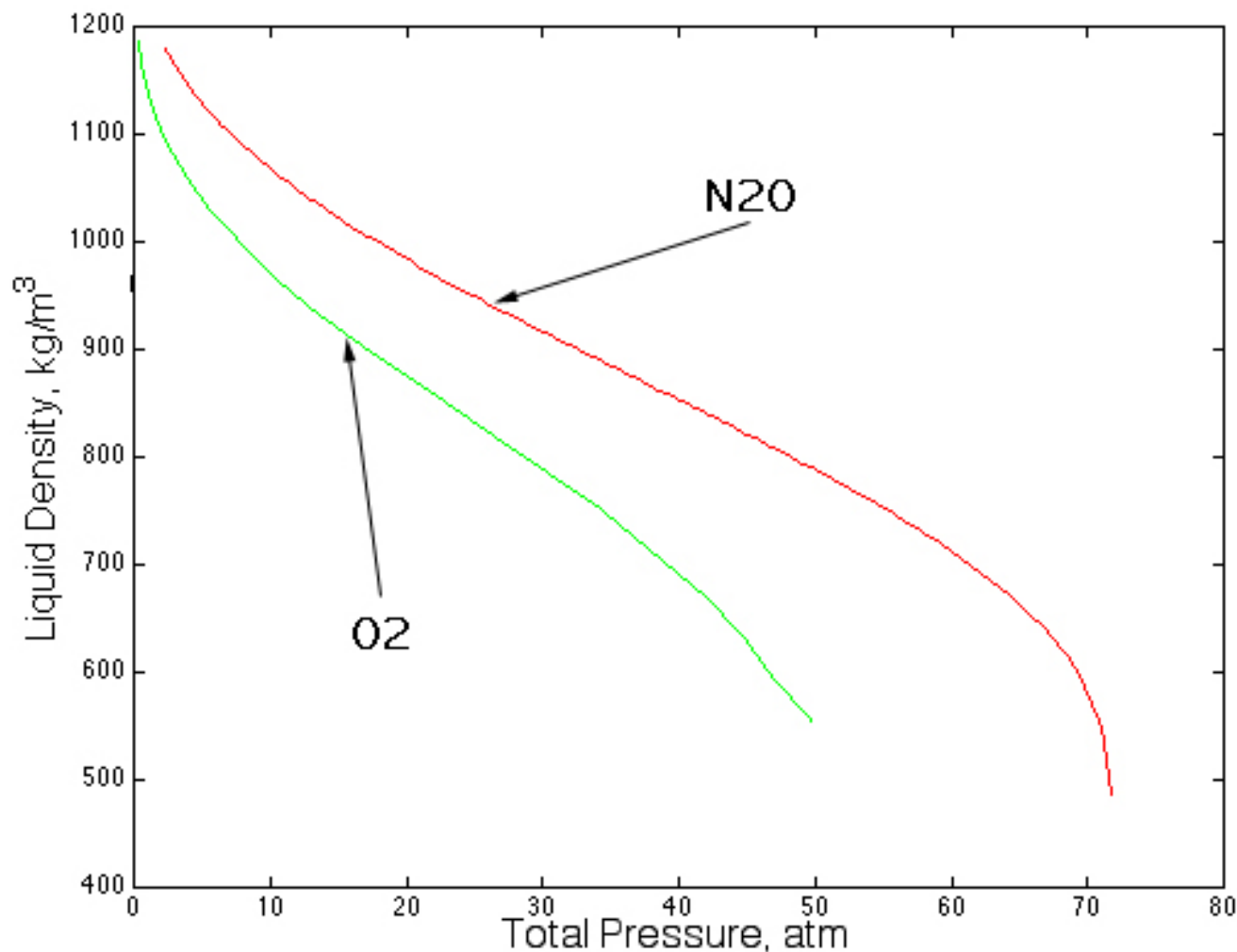
## Saturation Pressure and Density Plots for N<sub>2</sub>O





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## Saturation Pressure vs Saturation Density for Popular Oxidizers



# Advanced Rocket Propulsion

## Specific Heats

- Specific heats are defined as

$$c_p = \left. \frac{\partial h}{\partial T} \right|_P$$

$$c_v = \left. \frac{\partial e}{\partial T} \right|_v$$

$c_p$  : Specific Heat @ Cons. Pressure

$c_v$  : Specific Heat @ Cons. Volume

$\gamma$  : Ratio of Specific Heats

- For an ideal gas  $h = h(T)$   $e = e(T)$

$$c_p = \frac{dh}{dT}$$

$$c_v = \frac{de}{dT}$$

$$c_p - c_v = R$$

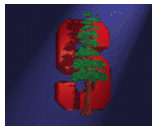
$$\gamma = \frac{c_p}{c_v}$$

- For a calorically perfect gas both specific heats are constant

$$c_p = \frac{N+2}{2} R$$

$$c_v = \frac{N}{2} R$$

- $N$  is the internal degrees of freedom (fully excited)



# Advanced Rocket Propulsion

## Specific Heats

- Monatomic gas:  $N=3$  (3 translational DoF)
- Diatomic gas
  - Vibrational modes NOT excited:  $N=5$  (+2 rotational DoF)
  - Vibrational modes fully excited:  $N=7$  (+2 vibrational DoF)

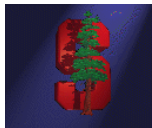
– For intermediate temperatures  $(c_v)_{vib} = R \left( \frac{\Theta_v}{T} \right)^2 \frac{e^{\Theta_v/T}}{(e^{\Theta_v/T} - 1)^2}$

$$\Theta_v \equiv \frac{h\nu}{k} = \text{Characteristic Temperature for Vibration}$$

$\nu$  = Frequency  
 $h$  : Planck's Cons.

- Data for diatomic molecules

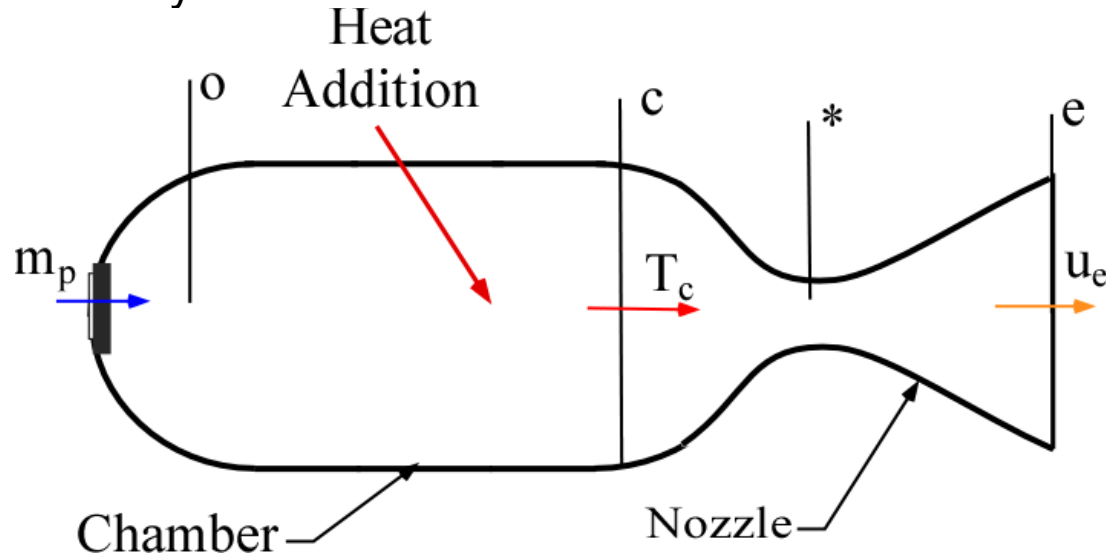
Characteristic Temperature	$N_2$ , K	$O_2$ , K
Rotational	2.9	2.1
Vibrational	3,390	2,270
Dissociation	113,000	59,500
Ionization	181,000	142,000



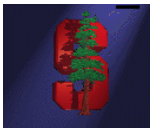
# Advanced Rocket Propulsion

## Thermal Rocket – General Concept

- In a thermal rocket the propellant molecules are thermalized by addition of heat in a chamber.
- This thermal energy (random motion of the molecules) is converted to the useful directional velocity needed for thrust in the nozzle.

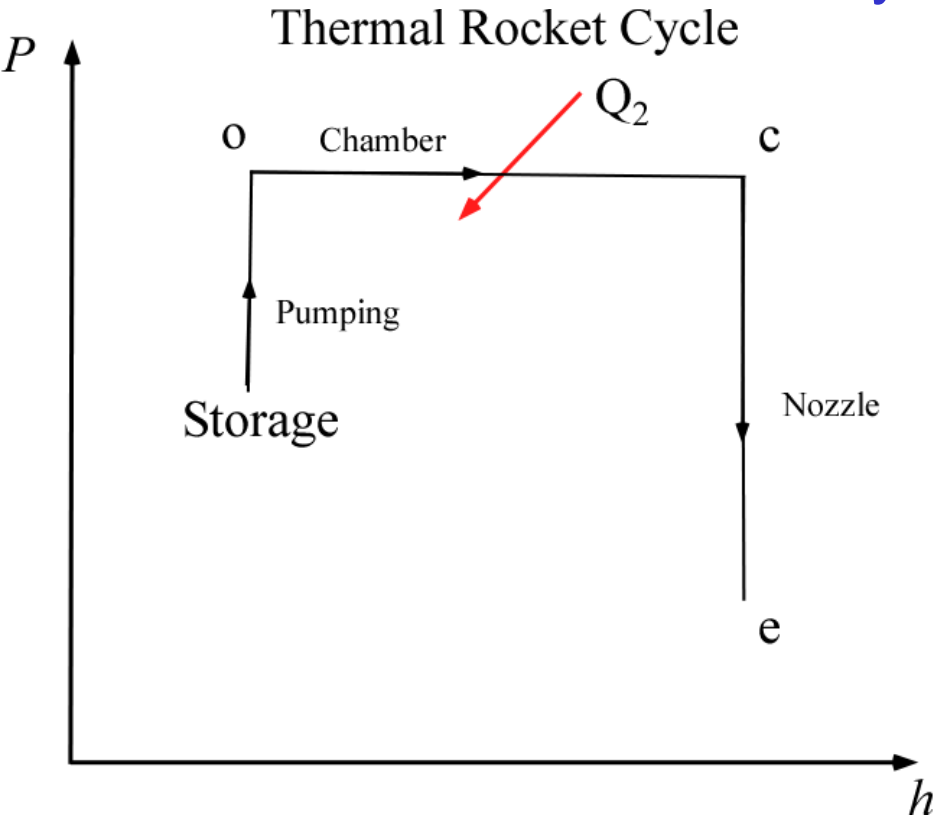


- The heat source varies
  - Nuclear energy: Thermonuclear rockets
  - Chemical bond energy: Chemical rockets
  - Electric energy: Resistojets and Arcjets
  - Thermal energy of the stored propellant: Cold gas thrusters



# Advanced Rocket Propulsion

## Thermal Rocket – Thermodynamic Process



- For monatomic gas:

$$u_e = 2.21 \sqrt{\frac{R_u T_{tc}}{Mw}}$$

- Note that this is not really a cycle since the propellant never returns to its original state
- The velocity at the nozzle exit,  $u_e$ , can be estimated using the conservation of energy along a streamline

$$h_{tc} - h_e = \frac{1}{2} u_e^2$$

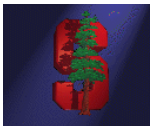
$$u_e = \sqrt{2(h_{tc} - h_e)}$$

- The maximum possible exit velocity (or  $I_{sp}$ ) is obtained for infinite expansion ( $h_e=0$ )

$$u_e = \sqrt{2h_{tc}}$$

- For calorically perfect gas

$$u_e = \sqrt{\frac{2\gamma}{\gamma-1}} \sqrt{\frac{R_u T_{tc}}{Mw}}$$



# Advanced Rocket Propulsion

## Thermal Rocket – Velocities in Monatomic Gas

- Lets compare this maximum velocity to the other fundamental velocities that can be defined in a monatomic gas

$$\text{Maximum Directional Velocity: } u_e = 2.21 \sqrt{\frac{R_u T_{tc}}{M_w}} \quad \text{Isentropic Speed of Sound: } a_s = 1.29 \sqrt{\frac{R_u T_{tc}}{M_w}}$$

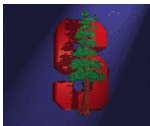
$$\text{Characteristic Velocity: } c^* = 1.38 \sqrt{\frac{R_u T_{tc}}{M_w}} \quad \text{Isothermal Speed of Sound: } a_T = \sqrt{\frac{R_u T_{tc}}{M_w}}$$

$$\text{Most Probable Molecular Speed (Maxwellian Distribution): } C_{MP} = 1.41 \sqrt{\frac{R_u T_{tc}}{M_w}}$$

$$\text{Average Molecular Speed (Maxwellian Distribution): } \bar{C} = 1.60 \sqrt{\frac{R_u T_{tc}}{M_w}}$$

$$\text{Square Average Molecular Speed: } \left(\bar{C}^2\right)^{1/2} = 1.72 \sqrt{\frac{R_u T_{tc}}{M_w}}$$

- Note that all of these speeds are of the same order (order of the average speed of the molecules in the gas)
- Similar results can be produced for other gamma values



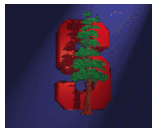
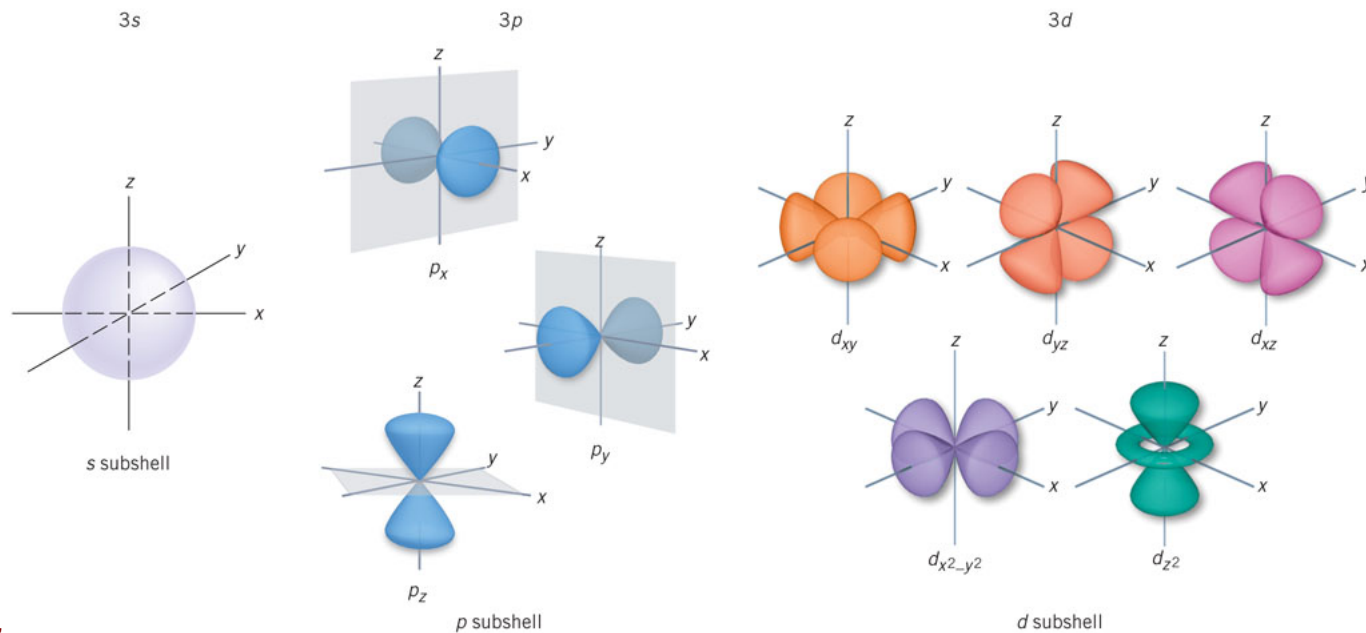
# Advanced Rocket Propulsion

## Review of Chemistry

- Atomic Model: Negatively charged electrons are orbiting around the positively charged nucleus
- Schrodinger's wave equation governs the size, number and shape of the orbitals

$$\left[ -\frac{h^2}{8\pi^2 m_e} \nabla^2 - \frac{Ze^2}{r} \right] \Psi = E\Psi$$

- Square norm of the wave function is a probability density function for the position or momentum of the electron
- Eigenvalue problem – Only discrete levels of energy is possible



# Advanced Rocket Propulsion

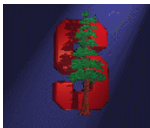
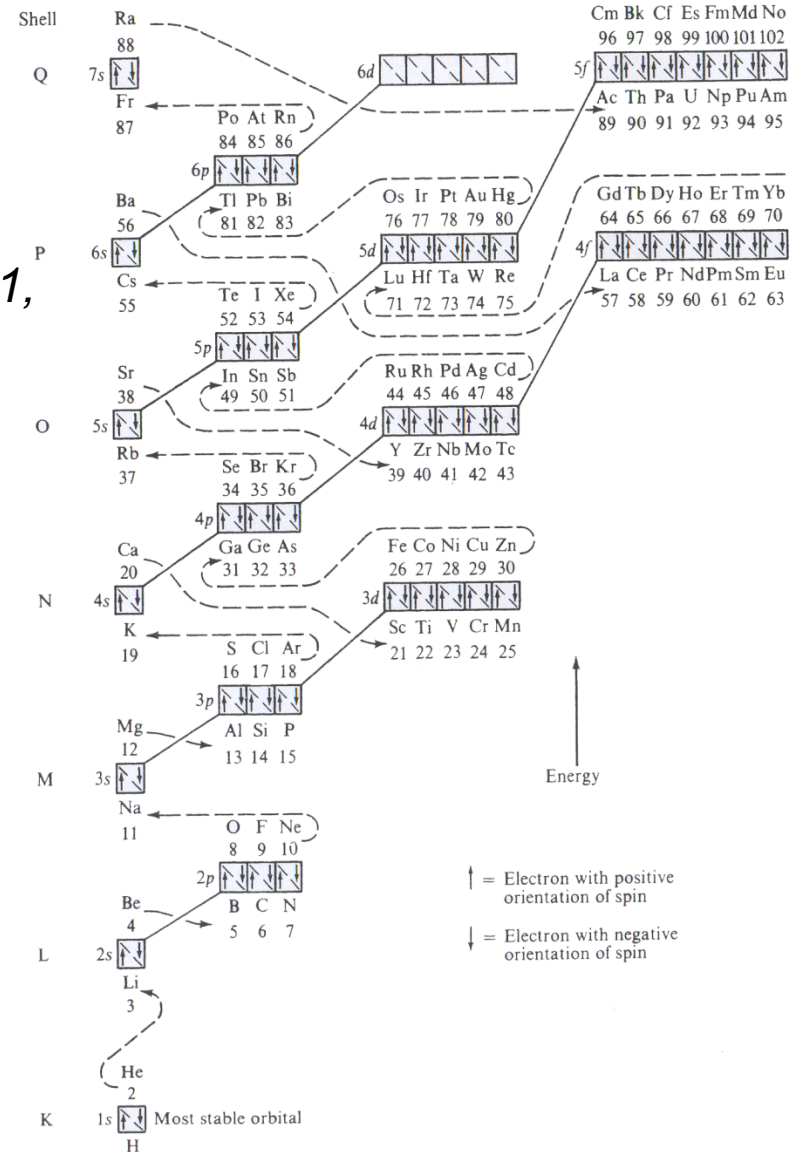
## Review of Chemistry – Energy Level Diagrams

### Atomic Structure (Quantum Mechanics)

- Principal quantum number:  $n=1, 2, 3 \dots$ 
  - General energy level of the shell
  - $n=1$  (K),  $n=2$  (L),  $n=3$  (M),...
- Angular momentum quantum number:  $l=0, 1, \dots, (n-1)$  [Orbital angular momentum]
  - Determines the shape of the orbitals
  - $l=0$  (s),  $l=1$  (p),  $l=2$ , (d), ...
- Magnetic quantum number:  $m_l: +l, +l-1, \dots, 0, \dots, -(l-1), l$ 
  - Determines the number of orbitals
- Electron Spin quantum number,  $m_s$ 
  - Each electron has:  $s=+\frac{1}{2}$  or  $-\frac{1}{2}$  [Intrinsic angular momentum the electron]

### Pauli Exclusion Principle:

“No two electrons in an atom may possess identical sets of values of the four quantum numbers  $n, l, m_l, m_s$ ”





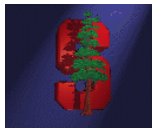
# Advanced Rocket Propulsion

## Review of Chemistry

- Quantum Mechanics Predicts:
  - Shells, subshells, orbitals
    - $n=1$ : K shell (2 electrons)
      - 1s subshell (1 orbital): 2 electrons
    - $n=2$ : L shell (8 electrons)
      - 2s subshell (1 orbital): 2 electrons
      - 2p subshell (3 orbitals): 6 electrons
    - $n=3$ : M shell
      - 3s subshell (1 orbital): 2 electrons
      - 3p subshell (3 orbitals): 6 electrons
      - 3d subshell: (5 orbitals): 10 electrons
  - Argononic structures: Completely full shells (Noble elements: He, Ne, Ar ...), Octets
  - Valance electrons: Electrons in the shell that is not completely filled

*n*: Principal Quantum Number

Reference on Chemistry: Linus Pauling, "General Chemistry"



# Advanced Rocket Propulsion

## Periodic Table of Elements

### Periodic Table of the Elements

1 H 1.008	2 He 4.003	3 Li 6.941	4 Be 9.012	5 Sc 44.96	6 Ti 47.87	7 V 50.94	8 Cr 52.00	9 Mn 54.94	10 Fe 55.85	11 Co 58.93	12 Ni 58.69	13 Cu 63.55	14 Zn 65.41	15 Ga 69.72	16 Ge 72.64	17 As 74.92	18 Se 78.96	19 Br 79.90	20 Kr 83.80															
11 Na 22.99	12 Mg 24.31	13 Al 26.98	14 Si 28.09	15 P 30.97	16 S 32.07	17 Cl 35.45	18 Ar 39.95	21 Y 88.91	22 Zr 91.22	23 Nb 92.91	24 Mo 95.94	25 Tc (97.9)	26 Ru 101.1	27 Rh 102.9	28 Pd 106.4	29 Ag 107.9	30 Cd 112.4	31 In 114.8	32 Sn 118.7	33 Sb 121.8	34 Te 127.6	35 I 126.9	36 Xe 131.3											
37 Rb 85.47	38 Sr 87.62	39 Y 88.91	40 Zr 91.22	41 Nb 92.91	42 Mo 95.94	43 Tc (97.9)	44 Ru 101.1	45 Rh 102.9	46 Pd 106.4	47 Ag 107.9	48 Cd 112.4	49 In 114.8	50 Sn 118.7	51 Sb 121.8	52 Te 127.6	53 I 126.9	54 Xe 131.3	55 Cs 132.9	56 Ba 137.3	57 La* 138.9	58 Ce 140.1	59 Pr 140.9	60 Nd 144.2	61 Pm (145)	62 Sm 150.4	63 Eu 152.0	64 Gd 157.3	65 Tb 158.9	66 Dy 162.5	67 Ho 164.9	68 Er 167.3	69 Tm 168.9	70 Yb 173.0	71 Lu 175.0
87 Fr (223)	88 Ra (226)	89 Ac~ (227)	104 Rf (261)	105 Db (262)	106 Sg (266)	107 Bh (264)	108 Hs (277)	109 Mt (268)	110 Ds (271)	111 Uuu (272)	112 Uub (277)	113 Uut -	114 Uuq -	115 Uup -	116 Uuh -	117 Uuq -	118 Uuo -	119 Uuh -	120 Uuo -															

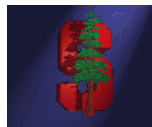


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\*Lanthanides

~Actinides

58 Ce 140.1	59 Pr 140.9	60 Nd 144.2	61 Pm (145)	62 Sm 150.4	63 Eu 152.0	64 Gd 157.3	65 Tb 158.9	66 Dy 162.5	67 Ho 164.9	68 Er 167.3	69 Tm 168.9	70 Yb 173.0	71 Lu 175.0
90 Th 232.0	91 Pa (231)	92 U (238)	93 Np (237)	94 Pu (244)	95 Am (243)	96 Cm (247)	97 Bk (247)	98 Cf (251)	99 Es (252)	100 Fm (257)	101 Md (258)	102 No (259)	103 Lr (262)



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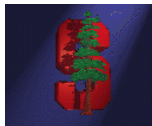


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# Advanced Rocket Propulsion

## Review of Chemistry

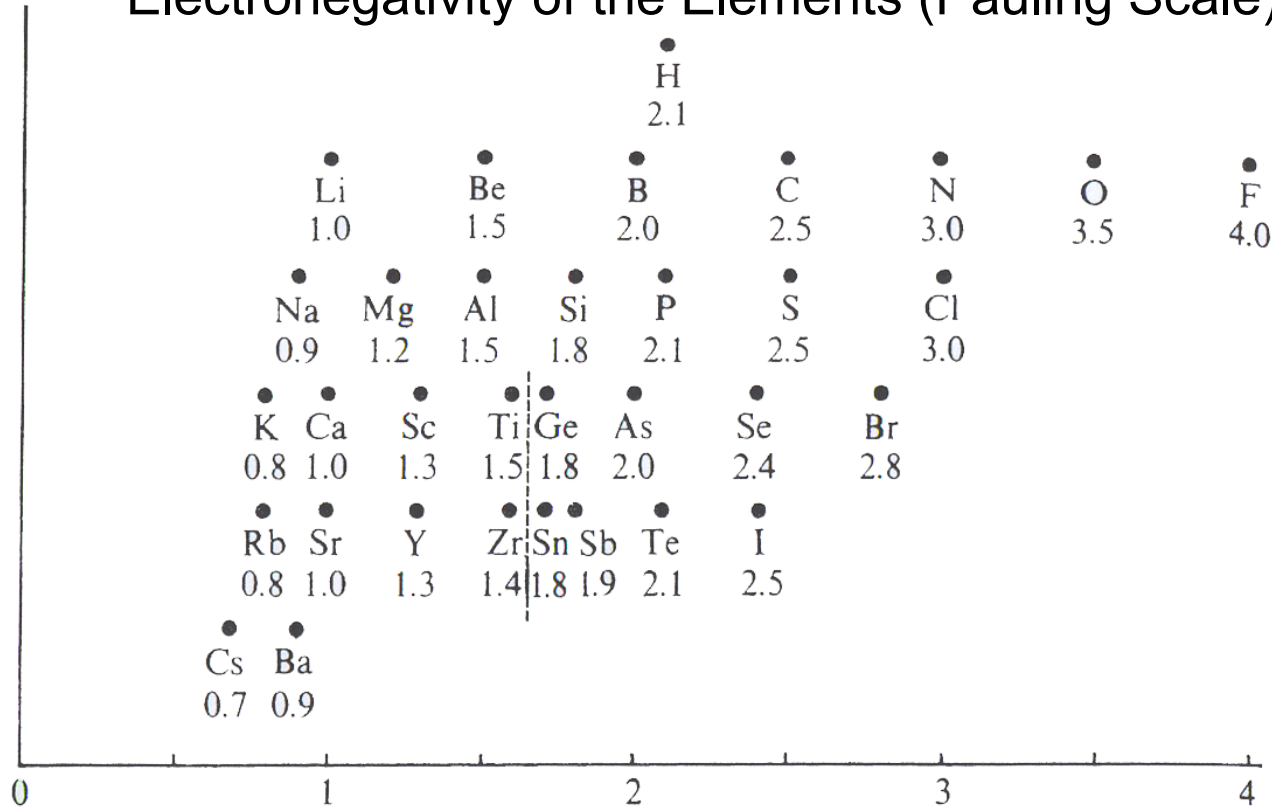
- Elements in the same group have closely related physical and chemical properties (same number of valence electrons)
- Elements in the periods have their valence electrons in the same shell
- Left side of the periodic table : Metals (Fuels)
  - High electric and thermal conductivity, metallic luster, malleable, ductile
- Right side of the periodic table : Nonmetals (Oxidizers)
- Metalloids in the middle: B, Si, Ge, As
- Chemical Bonds:
  - Octet rule: Filled shell rule
    - Share or gain electrons to fill their shells to the Argononic structures
  - Covalent Bonds: Share pairs of electrons (H-C)
  - Ionic Bonds: Take or give electrons (Li<sup>+</sup>Cl<sup>-</sup>)
- Electronegativity: Affinity of an atom to an electron
  - The difference in electronegativity determines the covalent/ionic nature of the bond (Upper right highly electronegative, lower left poorly electronegative)
- Strained bonds (C<sub>3</sub>H<sub>6</sub>, C<sub>2</sub>H<sub>2</sub>) and Resonance Structures (N<sub>2</sub>O)



# Advanced Rocket Propulsion

## Review of Chemistry – Electronegativity of the Elements

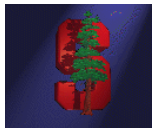
Electronegativity of the Elements (Pauling Scale)



Allred-Rochow Scale Electronegativity Coefficients: 
$$\chi = \frac{3590(Z_{eff} - 0.35)}{r_{cov}^2} + 0.744$$

$Z_{eff}$  : Effective Atomic Number

$r_{cov}$  : Covalent Radius



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# Advanced Rocket Propulsion

## Review of Chemistry – Electronegativity and Bonding

Influence of Electronegativity on Chemical Bonding

