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NASA-GLENN CHEMICAL EQUILIBRIUM PROGRAM CEA2, MAY 21, 2004
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 REFS: NASA RP-1311, PART I, 1994 AND NASA RP-1311, PART II, 1996

problem
 rocket fac ac/at=3.0 tcest,k=3800
 p,bar=100,
 sup,ae/at=100000,
 react
 fuel=H2 moles=2.0 t,k=298.15
 fuel=O2 moles=1.0 t,k=298.15
 end

OPTIONS: TP=F HP=F SP=F TV=F UV=F SV=F DETN=F SHOCK=F REFL=F INCD=F
 RKT=T FROZ=F EQL=T IONS=F SIUNIT=T DEBUGF=F SHKDBG=F DETDBG=F TRNSPT=F

TRACE= 0.00E+00 S/R= 0.000000E+00 H/R= 0.000000E+00 U/R= 0.000000E+00

Pc,BAR = 100.000000

Pc/P =

SUBSONIC AREA RATIOS =

SUPERSONIC AREA RATIOS =100000.0000

NFZ= 1 Mdot/Ac= 0.000000E+00 Ac/At= 3.000000E+00

REACTANT	MOLES	(ENERGY/R),K	TEMP,K	DENSITY
EXPLODED FORMULA				
F: H2	2.000000	-0.326752E-06	298.15	0.0000
H	2.00000			
F: O2	1.000000	-0.154035E-05	298.15	0.0000
O	2.00000			

SPECIES BEING CONSIDERED IN THIS SYSTEM
 (CONDENSED PHASE MAY HAVE NAME LISTED SEVERAL TIMES)
 LAST thermo.inp UPDATE: 9/09/04

g 6/97 *H	g 4/02 HO2	tpis78 *H2
g 8/89 H2O	g 6/99 H2O2	g 5/97 *O
g 4/02 *OH	tpis89 *O2	g 8/01 O3
g11/99 H2O(cr)	g 8/01 H2O(L)	g 8/01 H2O(L)

O/F = 0.000000

ENTHALPY	EFFECTIVE FUEL	EFFECTIVE OXIDANT	MIXTURE
(KG-MOL)(K)/KG	h(2)/R	h(1)/R	h0/R
	-0.60888815E-07	0.00000000E+00	-0.60888815E-07

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KG-FORM.WT./KG	bi(2)	bi(1)	b0i
*H	0.11101687E+00	0.00000000E+00	0.11101687E+00
*O	0.55508435E-01	0.00000000E+00	0.55508435E-01

POINT ITN	T	H	O
1 8	3730.278	-9.575	-14.967
2 2	3723.353	-9.593	-14.984
Pinf/Pt = 1.730122			
3 3	3542.876	-9.813	-15.178
Pinf/Pt = 1.727754			
3 2	3543.310	-9.812	-15.178
4 2	3716.555	-9.601	-14.991
4 2	3715.486	-9.602	-14.992
4 2	3715.394	-9.602	-14.992
2 2	3726.907	-9.584	-14.976
Pinf/Pt = 1.730186			
3 3	3546.007	-9.804	-15.170
Pinf/Pt = 1.727819			
3 2	3546.443	-9.803	-15.169
4 2	3720.093	-9.592	-14.983
4 2	3719.022	-9.593	-14.984
4 2	3718.930	-9.593	-14.984
END OF CHAMBER ITERATIONS			
4 9	219.523	-70.569	-26.937
4 3	306.519	-50.623	-27.675

DERIVATIVE MATRIX SINGULAR (EQLBRM)

SINGULAR MATRIX, ITERATION 1 VARIABLE 2(EQLBRM)

4	10	306.530	-51.978	-24.960
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THEORETICAL ROCKET PERFORMANCE ASSUMING EQUILIBRIUM

COMPOSITION DURING EXPANSION FROM FINITE AREA COMBUSTOR

Pin = 1450.4 PSIA
 Ac/At = 3.0000 Pinj/Pinf = 1.022603
 CASE =

	REACTANT	MOLES	ENERGY KJ/KG-MOL	TEMP K
FUEL	H2	2.0000000	0.000	298.150
FUEL	O2	1.0000000	0.000	298.150

O/F= 0.00000 %FUEL=100.00000 R,EQ.RATIO= 1.00000 PHI,EQ.RATIO= 0.00000

	INJECTOR	COMB END	THROAT	EXIT
Pinj/P	1.0000	1.0468	1.7669	8245875.
P, BAR	100.00	95.533	56.597	0.00001
T, K	3730.28	3718.93	3546.44	306.53
RHO, KG/CU M	5.1130 0	4.9007 0	3.0842 0	8.5722-6
H, KJ/KG	0.00000	-45.579	-1035.77	-13407.7
U, KJ/KG	-1955.80	-1994.96	-2870.84	-13549.2

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G, KJ/KG	-58737.3	-58647.8	-56920.0	-18238.0
S, KJ/(KG)(K)	15.7461	15.7578	15.7578	15.7578
M, (1/n)	15.858	15.862	16.069	18.015
(dLV/dLP)t	-1.04715	-1.04708	-1.04248	-1.00000
(dLV/dLT)p	1.7849	1.7862	1.7446	1.0000
Cp, KJ/(KG)(K)	10.1792	10.2076	10.0551	1.8665
GAMMAS	1.1324	1.1322	1.1289	1.3285
SON VEL,M/SEC	1488.2	1485.6	1439.3	433.5
MACH NUMBER	0.000	0.203	1.000	11.945

PERFORMANCE PARAMETERS

Ae/At	3.0001	1.0000	100000.0
CSTAR, M/SEC	2203.0	2203.0	2203.0
CF	0.1371	0.6533	2.3506
Ivac, M/SEC	6758.5	2714.3	5205.7
Isp, M/SEC	301.9	1439.3	5178.4

MOLE FRACTIONS

*H	0.03354	0.03351	0.02917	0.00000
HO2	0.00027	0.00026	0.00018	0.00000
*H2	0.12070	0.12052	0.11189	0.00000
H2O	0.68966	0.69025	0.71805	1.00000
H2O2	0.00005	0.00005	0.00003	0.00000
*O	0.01572	0.01568	0.01331	0.00000
*OH	0.10590	0.10552	0.09457	0.00000
*O2	0.03417	0.03419	0.03278	0.00000

* THERMODYNAMIC PROPERTIES FITTED TO 20000.K

PRODUCTS WHICH WERE CONSIDERED BUT WHOSE MOLE FRACTIONS
WERE LESS THAN 5.000000E-06 FOR ALL ASSIGNED CONDITIONS

O3	H2O(cr)	H2O(L)
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